



Wing lift= 674.3621 N  
 Wing lift tot= 674.3621 N  
 Wing drag= 40.8007 N  
 Wing drag ex= 6.1201 N  
 Wing drag tot= 46.9207 N  
 Lines drag= 50.1460 N (risers inc)  
 Pilot drag= 19.0172 N  
 Total drag= 116.0839 N  
 Lift to Drag ratios:  
 Wing Cl/Cd= 16.5282  
 $Cl^*C_{le} / (Cd^*C_{de}) = 14.3724$   
 $L / (D + D_{extra}) = 14.3724$   
 $L / (\text{Drag total}) = 5.8093$

# COMET